

FGM implementation in Ansys Fluent 15.0

Application to Sandia flame D

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Case description

This tutorial explains how to correctly implement the Fluent 15.0 built in Flamelet Generated Manifolds method and be able to simulate premixed flames. In particular this tutorial focuses on the simulation of the Sandia flame D.

The jet fluid is a mixture of three parts air and one part CH₄ by volume. The mixing rates are high enough that these flames burn as diffusion flames, with a single reaction zone near the stoichiometric mixture fraction and no indication of significant premixed reaction in the fuel-rich CH₄/air mixtures. Flame D operates at a Reynolds number of 22400. The pilot is a lean mixture of C₂H₂, H₂, air, CO₂, and N₂ with the equivalence ratio of 0.77.

Step-by-step guide to the simulation

Files needed

For the simulation you will need the mesh file “sandia-d.msh”, the thermal properties database “thermo.db” and the chemical mechanism file “gri-3.0.che”.

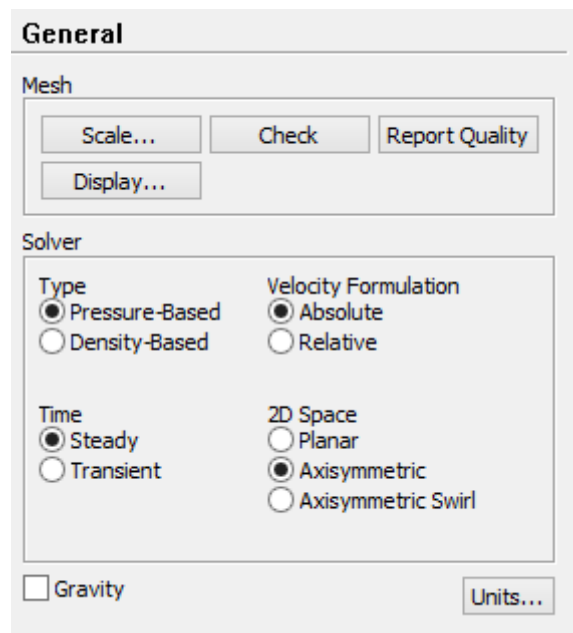
Starting of Fluent 15.0

In the Fluent launcher choose a “2D” dimension and leave the “Double Precision” checkbox unchecked. In “Working directory” browse the working path that you are using for the simulation.

Mesh import

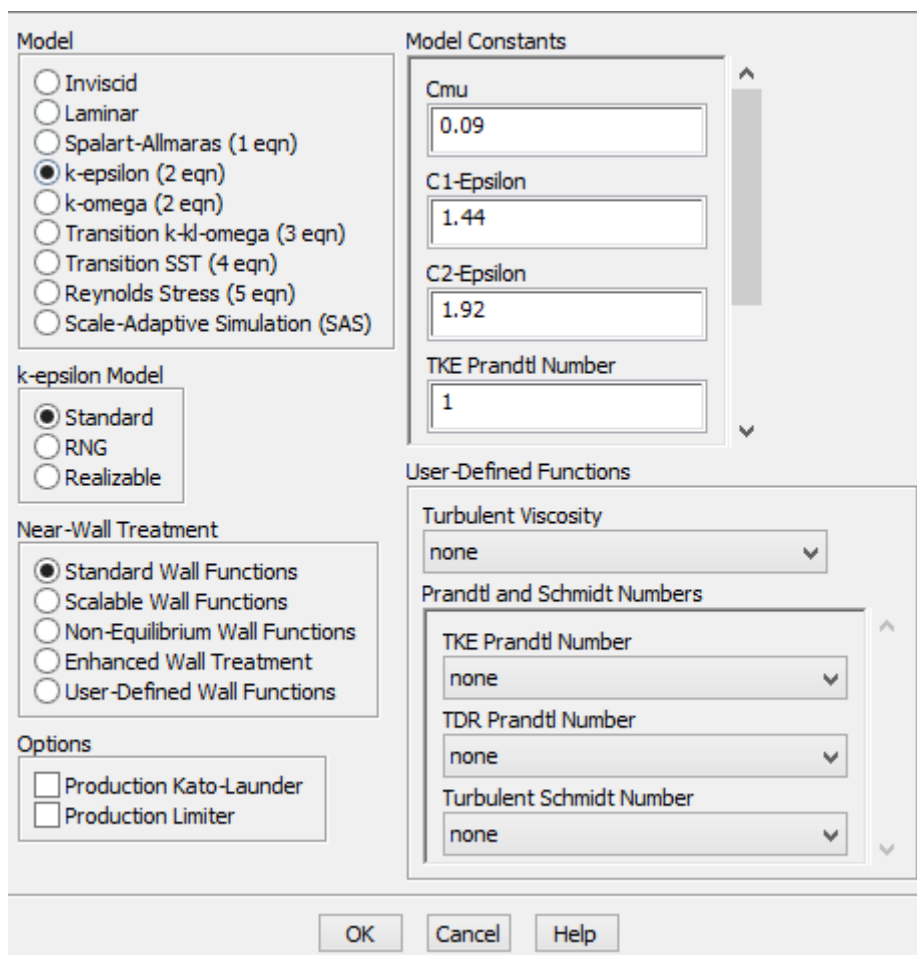
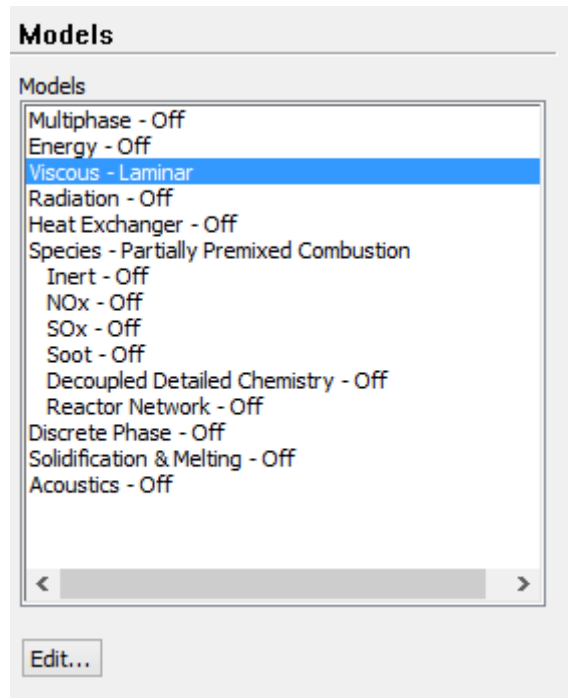
When Fluent is opened import the mesh by clicking on File→Read→Mesh...

Under the “Solution Setup” tree, in the “General” branch choose “Axisymmetric” under 2D space.



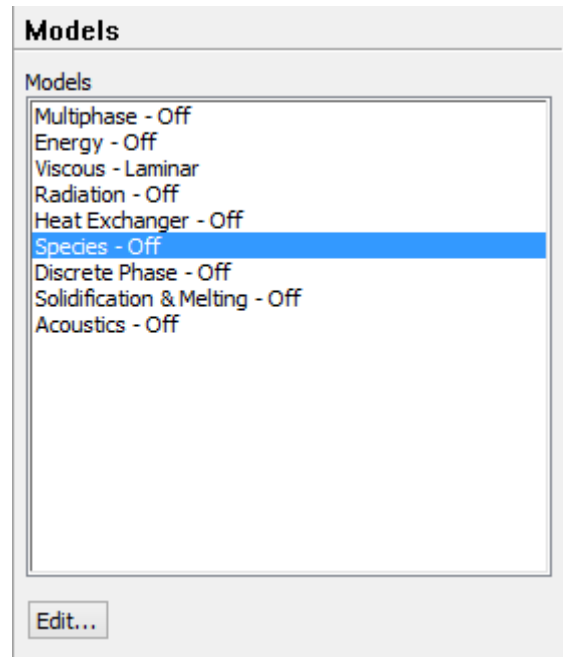
Turbulence model setup

Select “Model” under the “Solution Setup” tree. Click on “Viscous-Laminar” and then click to “Edit...” then check the “k-e (2 equation)” checkbox and leave all the parameters as default.

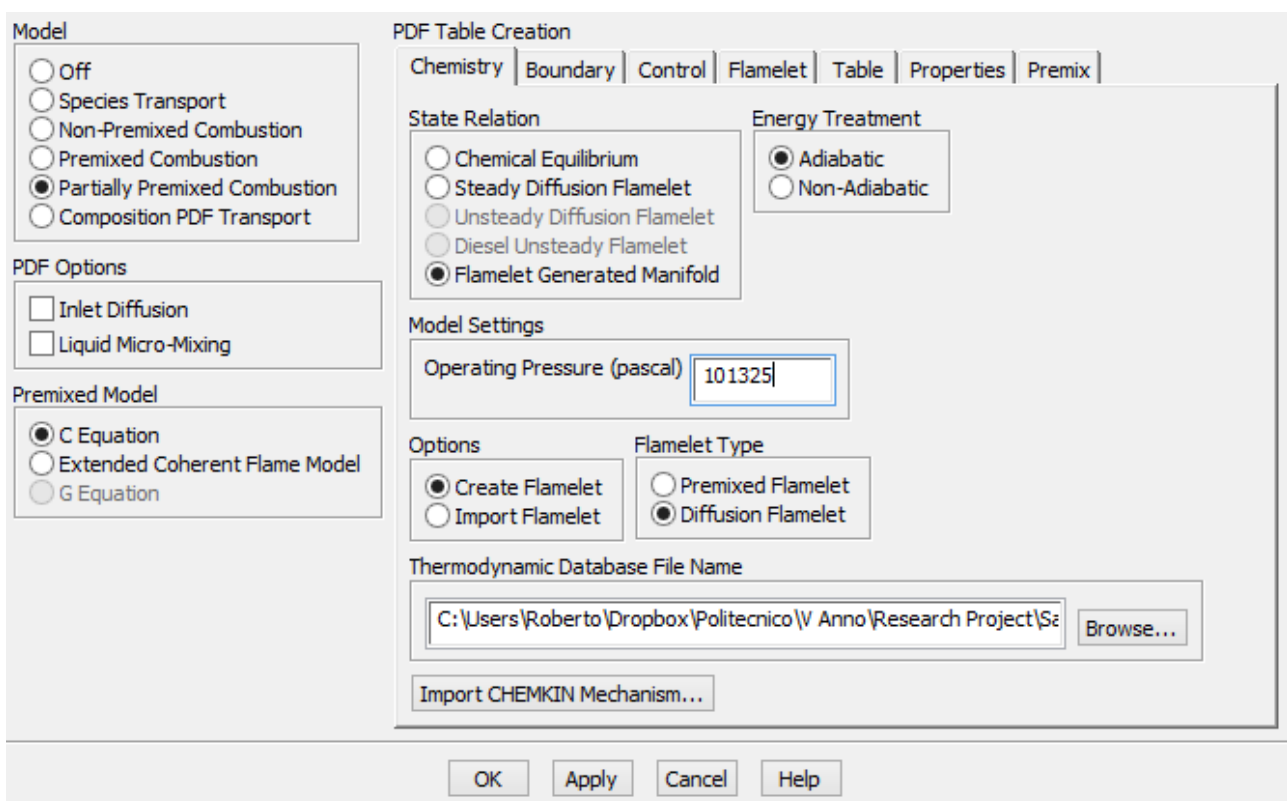


Set up of the Flamelet Generated Manifold model

Under the “Model” branch select “Species” and click on “Edit...”



Now check “Partially Premixed Combustion”. Set the operating pressure to 101325 Pa. Under “Options” select “Create Flamelet” and under “Flamelet Type” choose “Diffusion Flamelet”



To import the Thermodynamic Database file click on “Browse...” and select the “thermo.db” file that you should keep in the working directory.

To import the chemical mechanism click on “Import CHEMKIN Mechanism...”

Under “Gas-Phase CHEMKIN Mechanism File” browse to the “gri-3.0.che” file that you should keep in the working directory.

Under “Gas-Phase Thermodynamic Database File” browse again to the “thermo.db” file.

Material Name

Gas-Phase CHEMKin

Gas-Phase CHEMKin Mechanism File

Gas-Phase Thermodynamic Database File

Import Surface CHEMKin Mechanism
 Import Transport Property Database

Now move to the “Boundary Tab” and set the value of the mole fractions of the species as seen in the following table. Remember to check “Mole Fraction” under “Specify species in”.

Species	FUEL	OXIDIZER
O2	0.1575	0.21
N2	0.5925	0.79
CH4	0.25	0

Set the fuel temperature to 294 K and the oxidizer one to 291 K.

Model

Off
 Species Transport
 Non-Premixed Combustion
 Premixed Combustion
 Partially Premixed Combustion
 Composition PDF Transport

PDF Options

Inlet Diffusion
 Liquid Micro-Mixing

Premixed Model

C Equation
 Extended Coherent Flame Model
 G Equation

PDF Table Creation

Chemistry | **Boundary** | Control | Flamelet | Table | Properties | Premix

Species	Fuel	Oxid
h2	<input type="text" value="0"/>	<input type="text" value="0"/>
h	<input type="text" value="0"/>	<input type="text" value="0"/>
o	<input type="text" value="0"/>	<input type="text" value="0"/>
o2	<input type="text" value="0.1575"/>	<input type="text" value="0.21"/>
oh	<input type="text" value="0"/>	<input type="text" value="0"/>

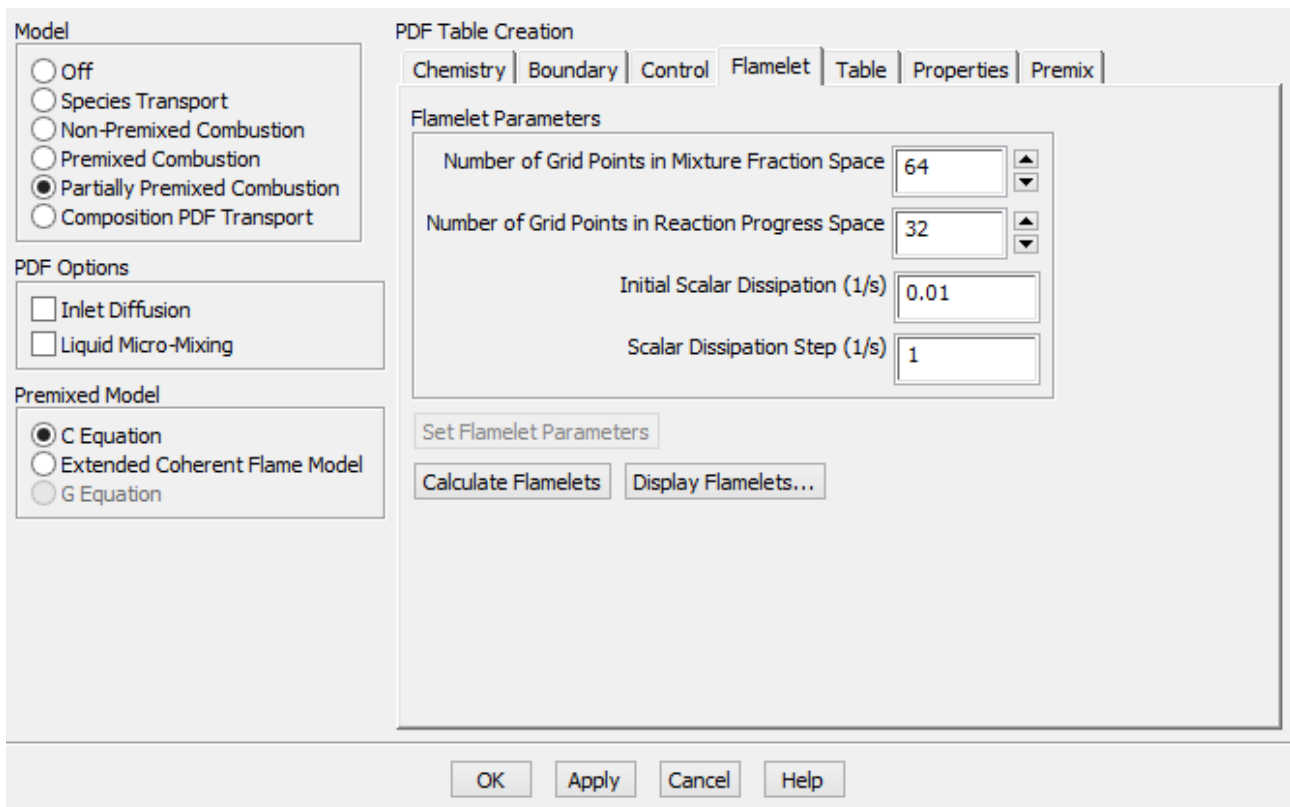
Temperature

Fuel (k)
Oxid (k)

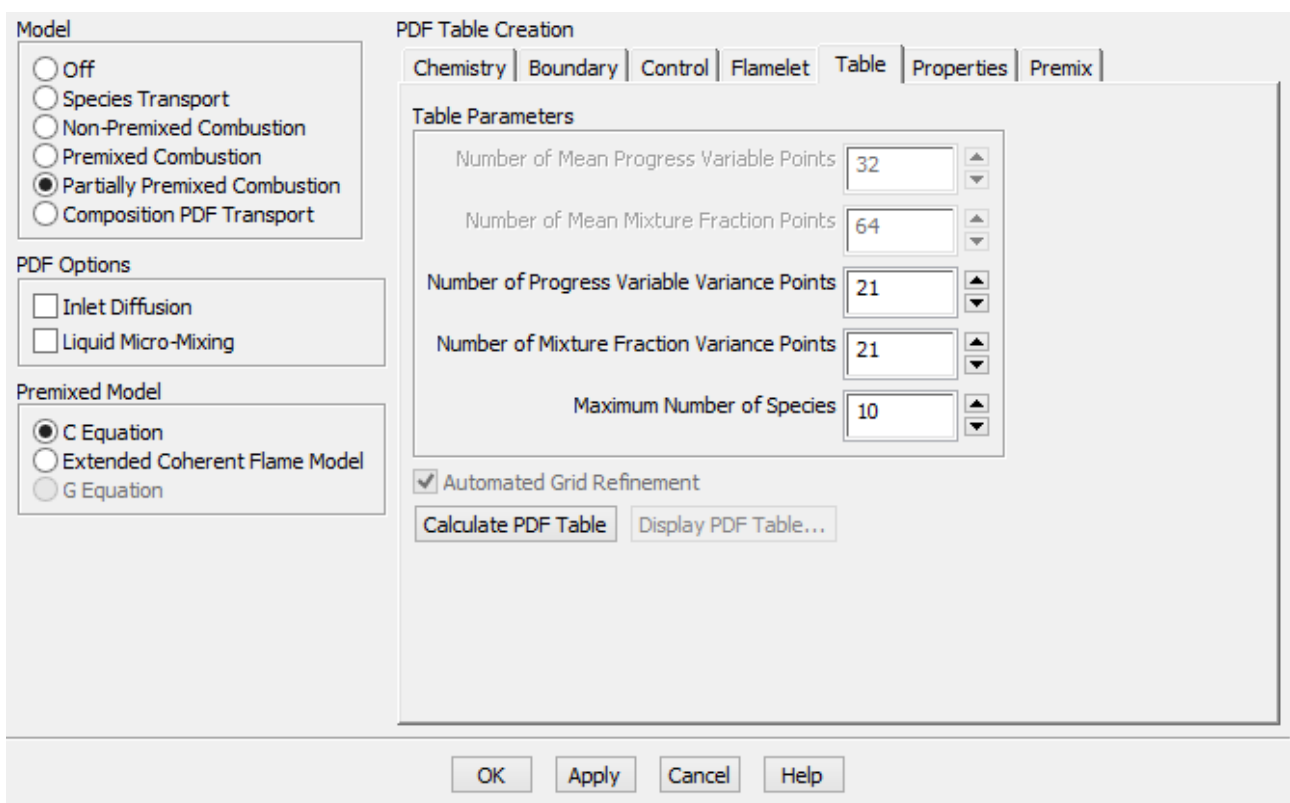
Specify Species in

Mass Fraction
 Mole Fraction

Leave the settings on the “Control” tab as default and move to the “Flamelet” tab. Modify the “Number of Grid Points in Mixture Fraction Space” to 64 and the “Number of Grid Points in Reaction Progress Space” to 32. Modify the “Scalar Dissipation Step” to 1. Now click on “Calculate Flamelets”



After the calculation is finished move to the “Table” tab. Select 21 for both “Number of Progress Variable Variance Points” and “Number of Mixture Fraction Variance Points”. Modify to 10 the “Maximum Number of Species”.



Click on “Calculate PDF Table”. When the calculations are finished move to the “Premix” Tab and under “Turbulence-Chemistry Interaction” select “Finite Rate”.

Boundary Conditions

Under the “Solution Setup” tree select the “Boundary Conditions” branch. Select “Coflow”, click “Edit...” and apply the momentum boundary conditions as shown in the figure below. Since “Coflow” is an oxidizer only velocity inlet the default species boundary conditions don’t need to be modified.

The screenshot shows the 'Boundary Conditions' dialog for the 'coflow' zone. The 'Zone Name' field contains 'coflow'. The 'Momentum' tab is selected. The 'Velocity Specification Method' is set to 'Magnitude, Normal to Boundary'. The 'Reference Frame' is set to 'Absolute'. The 'Velocity Magnitude (m/s)' is set to 0.9 with a 'constant' dropdown. The 'Supersonic/Initial Gauge Pressure (pascal)' is set to 0 with a 'constant' dropdown. The 'Turbulence' section is expanded, showing the 'Specification Method' set to 'Intensity and Viscosity Ratio'. The 'Turbulent Intensity (%)' is set to 10 and the 'Turbulent Viscosity Ratio' is set to 10. Both turbulence fields have a 'P' icon. At the bottom are 'OK', 'Cancel', and 'Help' buttons.

Select “Jet”, click “Edit...” and apply the momentum boundary conditions as shown in the figure below.

The screenshot shows the 'Boundary Conditions' dialog for the 'jet' zone. The 'Zone Name' field contains 'jet'. The 'Momentum' tab is selected. The 'Velocity Specification Method' is set to 'Magnitude, Normal to Boundary'. The 'Reference Frame' is set to 'Absolute'. The 'Velocity Magnitude (m/s)' is set to 49.6 with a 'constant' dropdown. The 'Supersonic/Initial Gauge Pressure (pascal)' is set to 0 with a 'constant' dropdown. The 'Turbulence' section is expanded, showing the 'Specification Method' set to 'Intensity and Hydraulic Diameter'. The 'Turbulent Intensity (%)' is set to 10 and the 'Hydraulic Diameter (m)' is set to 0.0072. Both turbulence fields have a 'P' icon. At the bottom are 'OK', 'Cancel', and 'Help' buttons.

Since “Jet” is the fuel inlet it is necessary to modify the species boundary condition changing the mixture fraction value to 1 (corresponding to 100% fuel and 0% oxidizer).

The screenshot shows a dialog box for the 'jet' zone. The 'Zone Name' field contains 'jet'. The 'Species' tab is selected. The 'Progress Variable' is set to 0, 'Progress Variable Variance' is 0, 'Mean Mixture Fraction' is 1, and 'Mixture Fraction Variance' is 0. All dropdown menus are set to 'constant'. The dialog has 'OK', 'Cancel', and 'Help' buttons at the bottom.

Select “Pilot”, click “Edit...” and apply the momentum boundary conditions as shown in the figure below.

The screenshot shows a dialog box for the 'pilot' zone. The 'Zone Name' field contains 'pilot'. The 'Momentum' tab is selected. The 'Velocity Specification Method' is 'Magnitude, Normal to Boundary', and the 'Reference Frame' is 'Absolute'. The 'Velocity Magnitude (m/s)' is 11.4, and 'Supersonic/Initial Gauge Pressure (pascal)' is 0. Both are set to 'constant'. Under the 'Turbulence' section, the 'Specification Method' is 'Intensity and Hydraulic Diameter', with 'Turbulent Intensity (%)' set to 10 and 'Hydraulic Diameter (m)' set to 0.0165. The dialog has 'OK', 'Cancel', and 'Help' buttons at the bottom.

Since “Pilot” is the pilot flame it is necessary to modify the species boundary condition changing the mixture fraction value to 0.2755 (corresponding to 27,55% fuel, 72.45% air) and the progress variable value to 1 (meaning that the mixture is already burnt).

Zone Name
pilot

Momentum | Thermal | Radiation | Species | DPM | Multiphase | UDS

Progress Variable	1	constant
Progress Variable Variance	0	constant
Mean Mixture Fraction	0.2755	constant
Mixture Fraction Variance	0	constant

OK Cancel Help

Solution methods

Now select “Solution Methods” branch under the “Solution” tree. Under “Spatial discretization” modify “Gradient” to “Green-Gauss Cell Based” and select a first order approximation for “Turbulent Kinetic Energy” and “Turbulent Dissipation Rate” while a second order approximation for all the other variables.

Solution controls

Select “Solution Controls” branch under the “Solution” tree. Modify the Pressure Under-Relaxation Factor to 0.7 and the Momentum one to 0.3. Leave all the other values to default.

Solution initialization

Select “Solution Initialization” branch under the “Solution” tree. Check “Standard Initialization”. Initialize the values as follows:

Axial Velocity: 0.9 m/s

Turbulent Kinetic Energy: 0.01215 m²/s²

Turbulent Dissipation Rate: 0.09 m²/s³

Progress Variable: 1

Leave all the other values to 0 then click on “Initialize”.

Monitors

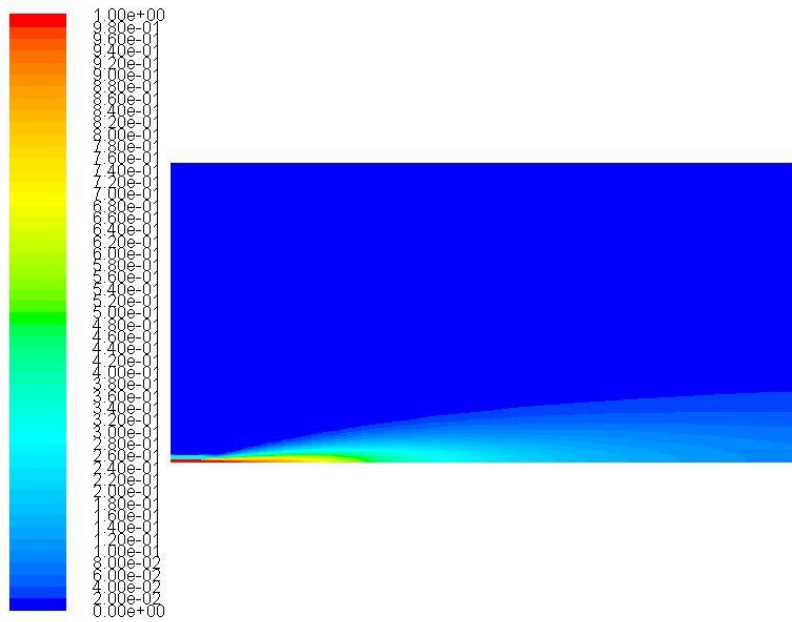
Select “Monitors” under the “Solution” tree. Select “Residuals” and click “Edit”. Then set “Convergence Criterion” to “none”.

Calculation

Select “Run Calculation” branch under the “Solution” tree. Modify the “Number of Iterations” to 2500. Click on “Calculate”.

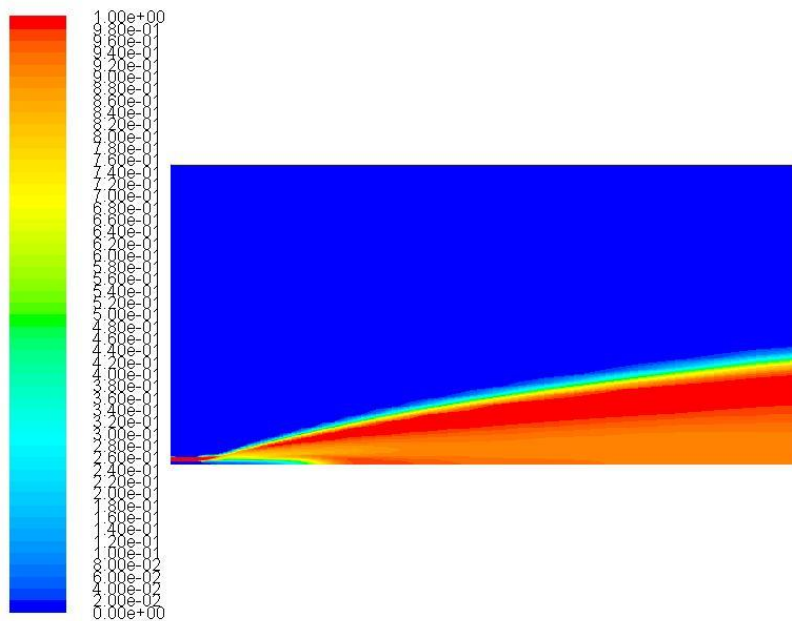
Solution

After the calculation is converged you should obtain a solution that is similar to the following:



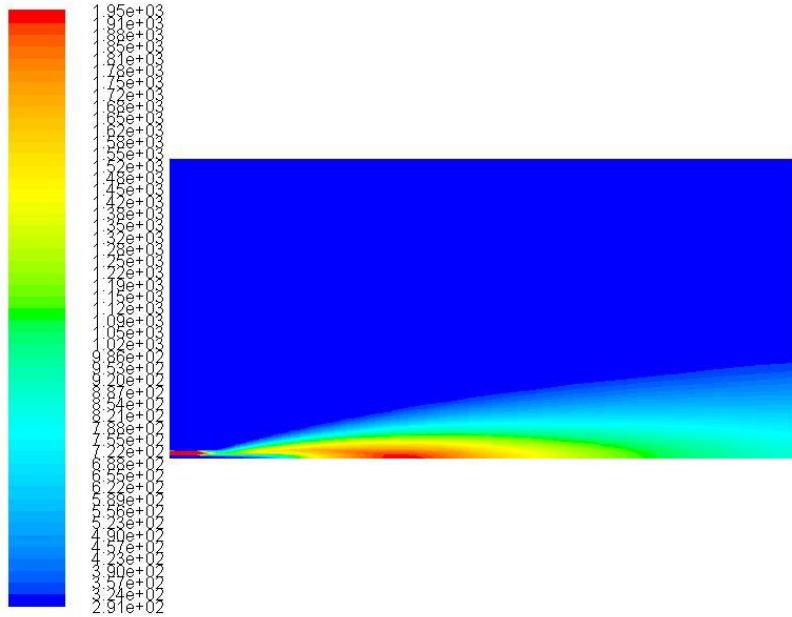
Contours of Mean Mixture Fraction

Feb 19, 2015
ANSYS Fluent 15.0 (axi, dp, pbns, pdf10, ske)



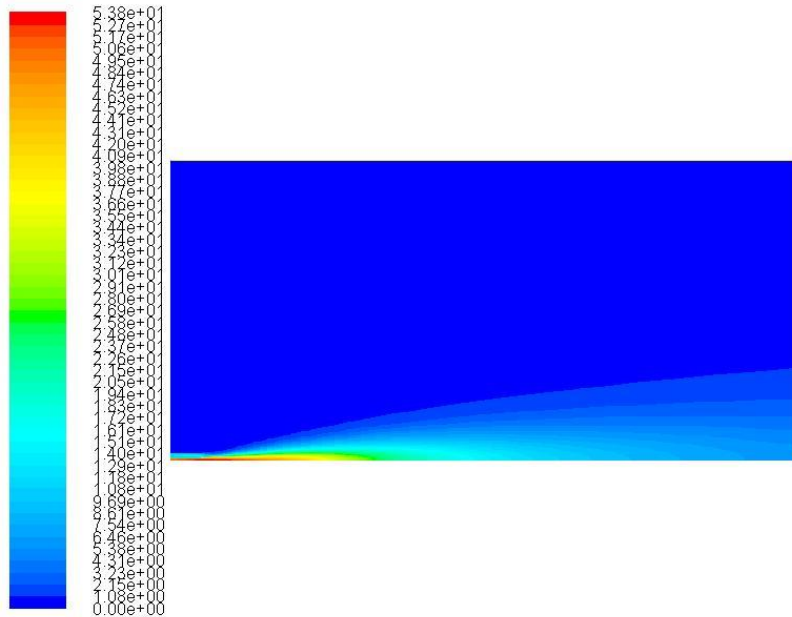
Contours of Progress Variable

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Contours of Static Temperature (k)

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Contours of Velocity Magnitude (m/s)

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