Airfoil Tools

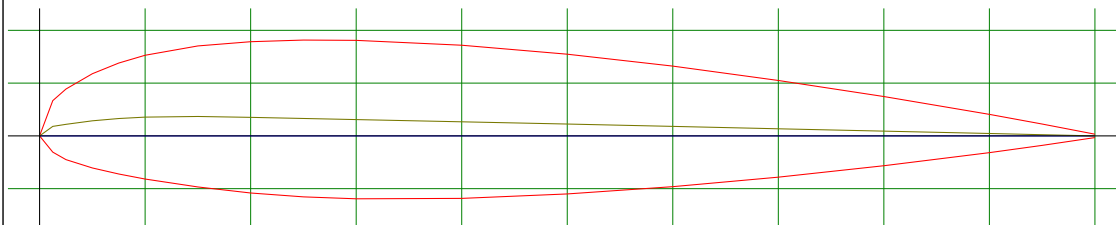Search 1636 airfoils   G+   Tweet   **Curtir** 1,9 mil

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NACA 23015 (naca23015-il)

NACA 23015 - NACA 23015 airfoil

←

Details

(naca23015-il) NACA 23015
NACA 23015 airfoil
Max thickness 15% at 30% chord.
Max camber 1.8% at 15% chord
Source **UIUC Airfoil Coordinates Database**
Source dat file
The dat file is in Selig format

Dat file

```
NACA 23015
1.0000    0.0016
0.9500    0.0112
0.9000    0.0204
0.8000    0.0373
0.7000    0.0525
0.6000    0.0661
0.5000    0.0774
```

Parser

No parser warnings

Send to airfoil plotter
Add to comparison
Lednicer format dat file
Selig format dat file

Similar airfoils

Airfoil		
NACA 63(2)-215 MOD B	Preview	Details
MH 91 14.98%	Preview	Details
MH 92 15.49%	Preview	Details
NACA 63(2)A-015	Preview	Details
KC-135 BL52.44 AIRFOIL	Preview	Details
MH 93 15.98%	Preview	Details
Gottingen 765 airfoil (recovered).	Preview	Details
MH 78 14.47%	Preview	Details
Falcon	Preview	Details
Smoothed ATR airfoil coordinates	Preview	Details

Polars for NACA 23015 (naca23015-il)

Plot	Airfoil	Reynolds #	Ncrit	Max Cl/Cd	Description	Source	
☑	naca23015-il	50,000	9	21 at α=4.75°	Mach=0 Ncrit=9	**Xfoil prediction**	Details
☐	naca23015-il	50,000	5	27.2 at α=6.25°	Mach=0 Ncrit=5	**Xfoil prediction**	Details
☑	naca23015-il	100,000	9	33.7 at α=6.75°	Mach=0 Ncrit=9	**Xfoil prediction**	Details
☐	naca23015-il	100,000	5	39.1 at α=7.5°	Mach=0 Ncrit=5	**Xfoil prediction**	Details
☑	naca23015-il	200,000	9	48.4 at α=9.5°	Mach=0 Ncrit=9	**Xfoil prediction**	Details
☐	naca23015-il	200,000	5	52.6 at α=8.25°	Mach=0 Ncrit=5	**Xfoil prediction**	Details
☑	naca23015-il	500,000	9	72 at α=10.5°	Mach=0 Ncrit=9	**Xfoil prediction**	Details
☐	naca23015-il	500,000	5	74.8 at α=8.75°	Mach=0 Ncrit=5	**Xfoil prediction**	Details
☑	naca23015-il	1,000,000	9	91 at α=10°	Mach=0 Ncrit=9	**Xfoil prediction**	Details
☐	naca23015-il	1,000,000	5	91.3 at α=9.25°	Mach=0 Ncrit=5	**Xfoil prediction**	Details

[Update plots]   **Reynolds number calculator**

Set Reynolds number and Ncrit range		Low	High
[Update Range]	Reynolds Number	50,000 ⌄	1,000,000 ⌄
	NCrit	7 ⌄	9 ⌄

Cl v Cd

Cl v Alpha

Cl/Cd v Alpha

Cd v Alpha

Cm v Alpha