

Airfoil Tools

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NACA 0015 (naca0015-il)

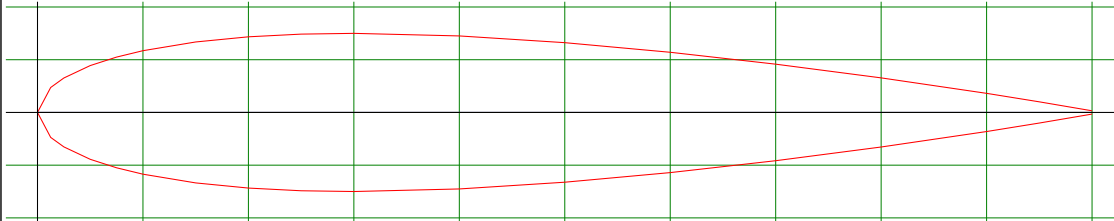
NACA 0015 - NACA 0015 airfoil



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Details

(naca0015-il) NACA 0015
 NACA 0015 airfoil
 Max thickness 15% at 30% chord.
 Max camber 0% at 0% chord
 Source [UIUC Airfoil Coordinates Database](#)
[Source dat file](#)
 The dat file is in Selig format

Dat file

```
NACA 0015
1.0000 0.00158
0.9500 0.01008
0.9000 0.01810
0.8000 0.03279
0.7000 0.04580
0.6000 0.05704
0.5000 0.06617
```

Parser

No parser warnings

- [Send to airfoil plotter](#)
- [Add to comparison](#)
- [Lednicer format dat file](#)
- [Selig format dat file](#)

Similar airfoils

FX 71-L-150/20 AIRFOIL	Preview	Details
FX 71-L-150/25 AIRFOIL	Preview	Details
FX 71-L-150/30 AIRFOIL	Preview	Details
FX 71-L-150/30 AIRFOIL	Preview	Details
WORTMANN FX L V-152 AIRFOIL	Preview	Details
NACA 63-015A AIRFOIL	Preview	Details
NACA 642-015 AIRFOIL	Preview	Details
Joukovsky f=0% t=15%	Preview	Details
E169 (14.4%)	Preview	Details
FX LIII-142 K 25 AIRFOIL	Preview	Details



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Polars for NACA 0015 (naca0015-il)

Plot	Airfoil	Reynolds #	Ncrit	Max Cl/Cd	Description	Source
<input checked="" type="checkbox"/>	naca0015-il	50,000	9	24.7 at $\alpha=6.25^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca0015-il	50,000	5	26.9 at $\alpha=6.25^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input checked="" type="checkbox"/>	naca0015-il	100,000	9	37.5 at $\alpha=6^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca0015-il	100,000	5	37.8 at $\alpha=6.25^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input checked="" type="checkbox"/>	naca0015-il	200,000	9	49.6 at $\alpha=6.25^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca0015-il	200,000	5	48.6 at $\alpha=7^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input checked="" type="checkbox"/>	naca0015-il	500,000	9	66.4 at $\alpha=7.5^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca0015-il	500,000	5	63.2 at $\alpha=8.75^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details
<input checked="" type="checkbox"/>	naca0015-il	1,000,000	9	77.9 at $\alpha=9^\circ$	Mach=0 Ncrit=9	Xfoil prediction Details
<input type="checkbox"/>	naca0015-il	1,000,000	5	75.1 at $\alpha=10.25^\circ$	Mach=0 Ncrit=5	Xfoil prediction Details

Update plots

[Reynolds number calculator](#)

Set Reynolds number and Ncrit range

Update Range

Reynolds Number

Low

50,000

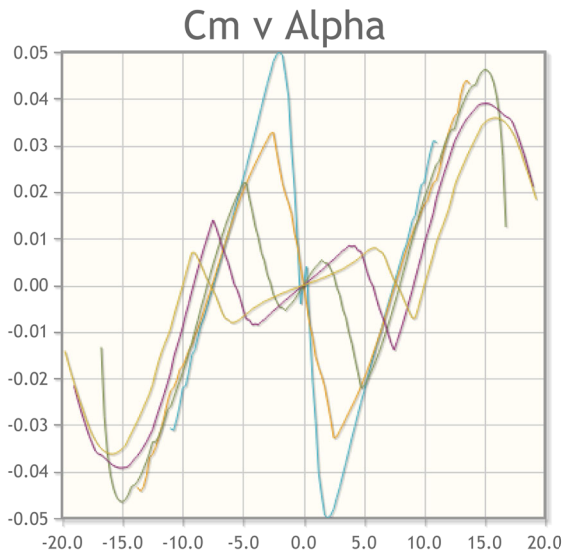
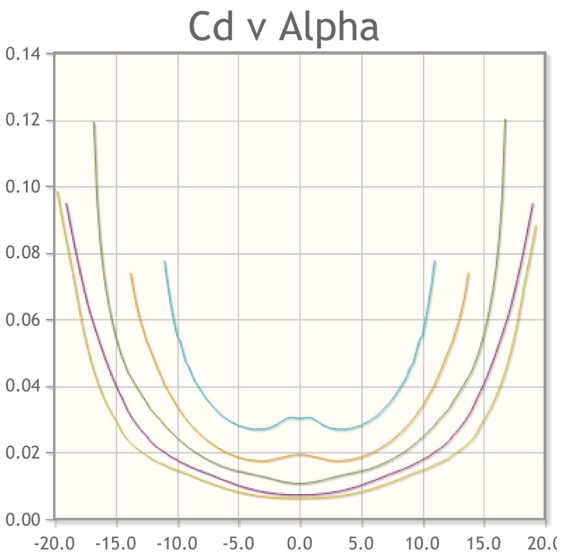
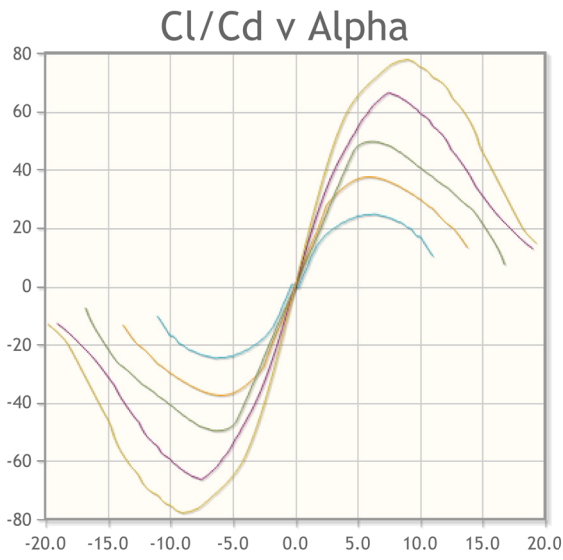
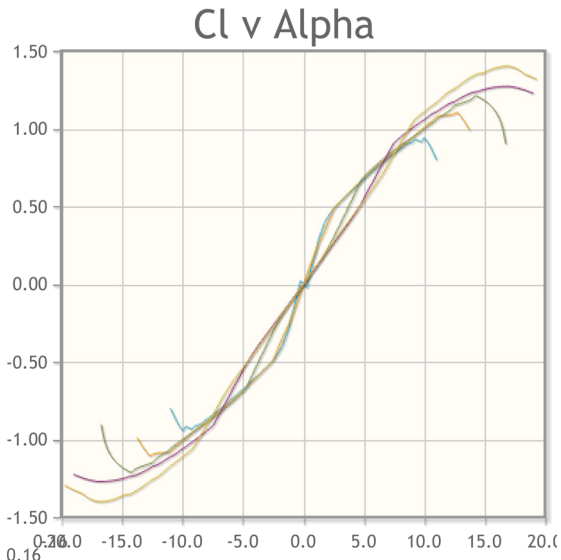
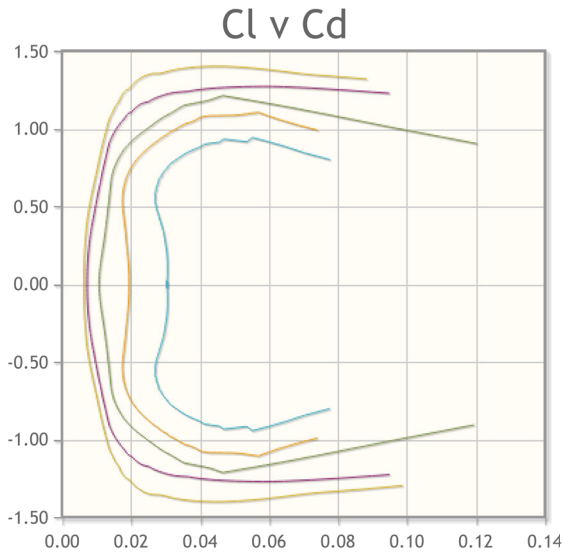
High

1,000,000

NCrit

7

9



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[NACA 0015 Aerofoil](#)

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